

DEPARTMENT OF TRANSPORTATION
FEDERAL AVIATION ADMINISTRATION

A18EA
Revision 10

2T-1A
2T-1A-1
2T-1A-2

September 27, 2000

TYPE CERTIFICATE DATA SHEET A18EA

This data sheet which is a part of Type Certificate No. A18EA prescribes conditions and limitations under which the product for which the Type Certificate was issued meets the airworthiness requirements of the Federal Aviation Regulations.

Type Certificate Holder: Great Lakes Aircraft Company, LLC
(John Duncan)
P.O. Box 9
Palmer Lake, Colorado 80133

Type Certificate Ownership Record:

A18EA Issued to Windward Aviation, Inc. on January 14, 1972

Reissued to Windward Aviation, Inc. on February 17, 1972

Reissued to Great Lakes Aircraft Company on September 5, 1972 to add Models 2T-1A-1, and 2T-1A-2.

Reissued to Great Lakes Aircraft Company (Russel Dean Franklin, Jr.) on September 6, 1979

Great Lakes Aircraft Company (Russel Dean Franklin, Jr) transferred ownership of Type Certificate A18EA to Chaparral Motors, Inc. on January 17, 1989.

Chaparral Motors, Inc. (John Duncan) changes to name and address on September 27, 2000.

I - Model 2T-1A, 1 or 2 POLB, Approved 1/14/72

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|-----------------------------------|--|
| Engine | American Cirrus Mark III |
| Fuel | 73 minimum octane aviation gas |
| Engine limits | Takeoff 2100 r.p.m. (100 hp.) Maximum Continuous 2100 r.p.m. (100 hp.) |
| Propeller and propeller limits | Fixed pitch wood or adjustable metal (Hamilton Standard 7020 hub, 25V2 blades) See NOTE 4 |
| Airspeed limits (CAS) | Maximum speed (S.L.) 100 m.p.h.. (95.5 knots) Landing speed 40 m.p.h. (34.8 knots) Cruising speed 90 m.p.h. (78.2 knots) |
| M.A.C. | 45.75 in. (L.E. of MAC is 10 in. forward of leading edge of lower wing) |

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| Maximum weight | 1618 lb. | | | | | |
| Minimum Crew | 1 (pilot) | | | | | |
| No. of seats | Two | | | | | |
| Maximum baggage | 13 lb. as 2-place 200 lb. as 1-place (Pay load includes 2 parachutes, 20 lb. each) | | | | | |
| Fuel capacity | 26 gal. (+60.5) | | | | | |
| Oil capacity | 8 qt. (+22.5) | | | | | |
| Control surface movements | Aileron | ($\pm 1^\circ$) | Up | 24° | Down | 22° |
| | Stabilizer | ($\pm 1/2^\circ$) | Up | 7-1/4° | Up | 1/4° |
| | Elevator | ($\pm 1^\circ$) | Up | 21° | Down | 29° |
| | Rudder | ($\pm 1^\circ$) | Left | 35° | Right | 35° |
| Serial Nos. eligible | 0501 and 0502 See NOTE 5 | | | | | |

II - Model 2T-1A-1, 1 or 2 POLB (Acrobatic Category) (See NOTE 6), Approved 12/17/73

| | | | | | | | | | |
|--------------------------------|--|--------------|------------------------|------------------------|------------------------|-------------|------------------------|--------------------|-----------------------|
| Engine | Lycoming O-320-E2A installed per Great Lakes Dwg. 50123; or Lycoming IO-360-B1F6 or AEIO-360-B1G6 installed per Great Lakes Dwg. 50148 | | | | | | | | |
| Fuel | Lycoming O-320-E2A, 80/87 minimum grade aviation fuel Lycoming IO-360-B1F6 and AEIO-360-B1G6, 91/96 minimum grade aviation fuel | | | | | | | | |
| Engine limits | <p>Lycoming O-320-E2A</p> <table> <tr> <td>Takeoff</td><td>2450 r.p.m. (140 hp.)</td></tr> <tr> <td>Maximum Continuous</td><td>2450 r.p.m. (140 hp.)</td></tr> </table> <p>Lycoming IO-360-B1F6 and AEIO-360-B1G6</p> <table> <tr> <td>Takeoff</td><td>2700 r.p.m. (180 hp.)</td></tr> <tr> <td>Maximum continuous</td><td>2700 r.p.m. (180 hp.)</td></tr> </table> | Takeoff | 2450 r.p.m. (140 hp.) | Maximum Continuous | 2450 r.p.m. (140 hp.) | Takeoff | 2700 r.p.m. (180 hp.) | Maximum continuous | 2700 r.p.m. (180 hp.) |
| Takeoff | 2450 r.p.m. (140 hp.) | | | | | | | | |
| Maximum Continuous | 2450 r.p.m. (140 hp.) | | | | | | | | |
| Takeoff | 2700 r.p.m. (180 hp.) | | | | | | | | |
| Maximum continuous | 2700 r.p.m. (180 hp.) | | | | | | | | |
| Propeller and propeller limits | <p>Lycoming O-320-E2A McCauley 1C160-EGM 7654 Diameter: Not over 76 in., not under 74.5 in. Static r.p.m. at maximum permissible throttle setting: Not over 2350 r.p.m., not under 2250 r.p.m. No additional tolerance permitted.</p> <p>Lycoming IO-360-B1F6 and AEIO-360-B1G6,</p> <p>(a) Hartzell HC-C2YK-4F/FC7666A-2 Diameter: 74.0 in. No reduction allowed Pitch settings at 30 in. sta.: low 12.5°, high 26.8°</p> <p>(b) Hartzell Spinner P/N 835-41</p> <p>(c) Hartzell Hydraulic Governor No. F6-31</p> | | | | | | | | |
| Airspeed limits (CAS) | <p><u>Landplane</u></p> <table> <tr> <td>Never exceed</td><td>142 m.p.h. (124 knots)</td></tr> <tr> <td>Max. structural cruise</td><td>120 m.p.h. (105 knots)</td></tr> <tr> <td>Maneuvering</td><td>120 m.p.h. (105 knots)</td></tr> </table> | Never exceed | 142 m.p.h. (124 knots) | Max. structural cruise | 120 m.p.h. (105 knots) | Maneuvering | 120 m.p.h. (105 knots) | | |
| Never exceed | 142 m.p.h. (124 knots) | | | | | | | | |
| Max. structural cruise | 120 m.p.h. (105 knots) | | | | | | | | |
| Maneuvering | 120 m.p.h. (105 knots) | | | | | | | | |
| M.A.C. | 46.0 in. (L.E. of MAC is Sta. 9.0) | | | | | | | | |

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| C.G. Range | (15.7) at 1600 lb. or less (18.9 to 23.0) at 1750 lb. Straight line variation between points given |
| Empty Wt. C.G. | None |
| Maximum weight | 1750 lb. |
| No. of seats | 2 (1 at 27.5, 1 at 61.5) |
| Maximum baggage | 40 lb. (+92.0) |
| Fuel capacity | 27.4 gal. (+26.7 gal. usable, one 26.0 gal. tank in top wing at +13.0 and one 1.4 gal. header tank in fuselage at +5.0). (Lycoming IO-360-B1F6 and AEIO-360-B1G6) See NOTE 1 for data on unusable fuel 26 gal. (26.0 gal. usable, one 26 gal. tank in top wing at +13.0) (Lycoming O-320-E2A) See NOTE 1 for data on unusable fuel |
| Oil capacity | 8 qt. (-24.0) (4 qt. usable) Lycoming IO-360-B1F6 and AEIO-360-B1G6 8 qt. (-24.0) (6 qt. usable) Lycoming O-320-E2A See NOTE 1 for data on oil system |
| Control surface movements | Aileron $(\pm 1^\circ)$ Up 24° Down 22° for both 2-aileron and 4-aileron configuration Stabilizer $(\pm 1/2^\circ)$ Up $7-1/4^\circ$ Up $1/4^\circ$ Elevator $(\pm 1^\circ)$ Up 21° Down 29° Rudder $(\pm 1^\circ)$ Left 35° Right 35° |
| Serial Nos. eligible | 0503 and 0699 See NOTE 5 |

I - Model 2T-1A-2, 1 or 2 POLB (Acrobatic Category) (See NOTE 6) Approved 7/25/74

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| Engine | Lycoming IO-360-B1F6 or AEIO-360-B1G6 installed per Great Lakes Dwg. 50148 or Lycoming O-320-E2A installed per Great Lakes Dwg. 50123 |
| Fuel | Lycoming IO-360-B1F6 and AEIO-360-B1G6, 91/96 minimum grade aviation fuel Lycoming O-320-E2A, 80/87 minimum grade aviation fuel |
| Engine limits | Lycoming IO-360-B1F6 and AEIO-360-B1G6 Takeoff 2700 r.p.m. (180 hp.) Maximum Continuous 2700 r.p.m. (180 hp.) Lycoming O-320-E2A Takeoff 2450 r.p.m. (140 hp.) Maximum continuous 2450 r.p.m. (140 hp.) |
| Propeller and propeller limits | Lycoming IO-360-B1F6 and AEIO-360-B1G6, (a) Hartzell HC-C2YK-4F/FC7666A-2 Diameter: 74.0 in. No reduction allowed Pitch settings at 30 in. sta.: low 12.5° , high 26.8° (b) Hartzell Spinner P/N 835-41 (c) Hartzell Hydraulic Governor No. F6-31 |

Lycoming O-320-E2A
McCauley 1C160-EGM 7654
Diameter: Not over 76 in., not under 74-5 in.
Static r.p.m. at maximum permissible throttle setting:
Not over 2350 r.p.m., not under 2250 r.p.m.
No additional tolerance permitted.

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|---------------------------|--|------------|-------------|-----------|--|
| Airspeed limits | <u>Landplane</u> | | | | |
| | Never exceed | 153 m.p.h. | (133 knots) | | |
| | Max. structural cruise | 120 m.p.h. | (105 knots) | | |
| | Maneuvering | 120 m.p.h. | (105 knots) | | |
| C. G. range | (+15.7) at 1600 lb. or less (+20.0) to (+23.8) at 1800 lb. Straight line variation between points given | | | | |
| Empty Wt. C.G. | None | | | | |
| Maximum weight | 1800 lb. | | | | |
| No. of seats | 2 (1 at 27.5, 1 at 61.5) | | | | |
| Maximum baggage | 40 lb. (+92.0) | | | | |
| Fuel capacity | 27.4 gal. (26.7 gal. usable, one 26.0 gal. tank in top wing at +13.0 and one 1.4 gal. header tank in fuselage at +5.0). (Lycoming IO- 360-B1F6 and AEIO-360-B1G6) See NOTE 1 for data on unusable fuel 26 gal. (26.0 gal. usable, one 26 gal. tank in top wing at +13.0) (Lycoming O-320-E2A) See NOTE 1 for data on unusable fuel | | | | |
| Oil capacity | 8 qt. (-24.0) (4 qt. usable) Lycoming IO-360-B1F6 and AEIO-360-B1G6 See NOTE 1 for data on oil system 8 qt. (-24.0) (6 qt. usable (Lycoming O-320-E2A) | | | | |
| Control surface movements | Aileron | (±1°) | Up 24° | Down 22° | |
| | Stabilizer | (±1/2°) | Up 7-1/4° | Up 1/4° | |
| | Elevator | (±1°) | Up 21° | Down 29° | |
| | Rudder | (±1°) | Left 35° | Right 35° | |
| Serial Nos. eligible | 0701 and up | | | | |

Data Pertinent to all Models

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| Datum | Fuselage Sta. (00) is center of most forward lateral fuselage tube. A pin is located on the lower side of landing gear streamline tube to indicate datum. |
| Leveling means | Upper Longeron at cockpit |
| Certification basis | Aeronautics Bulletin No. 7-A dated October 1, 1934; FAR 23 dated February 1, 1967, Amendments 23-1 through 23-7 for powerplant installation; Type Certificate A18EA issued January 14, 1972. Great Lakes Company obtained Exemption No. 1163 which granted an exemption from Section 21.17 of the FAR to permit the issuance of the Type Certificate incorporating the type design portion of ATC-228. |
| Production basis | None. Prior to original certification of each aircraft manufactured subsequent to May 5, 1982, an FAA representative must perform a |

detailed inspection for workmanship, materials, and conformity with the approved technical data, and a check of the flight characteristics.

Equipment

The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.

In addition, the following items of equipment are required:

1. FAA approved Airplane Flight Manual as follows:
 - a. Serial Nos. 0501 and 0502, approval date July 13, 1973, or later FAA approved revision as required by STC SA941CE.
 - b. Serial Nos. 0503 through 0699; approval date Nov. 9, 1973, or later FAA approved revision.
 - c. Serial Nos. 0701 and up; approval date July 25, 1974, or later FAA approved revision.

NOTE 1. Current weight and balance report together with list of equipment included in certificated empty weight must be provided for each aircraft at time of original certification. The certificated empty weight and corresponding center of gravity location must include unusable fuel of 0.0 lb. at (+13.0) and system oil of 1.0 lb. at (-4.0) for airplanes with Lycoming O-320-E2A engines. Certificated empty weight and corresponding center of gravity location must include unusable fuel of 4.0 lb. at (+5.0) and system oil of 1.0 lb. at (-4.0) for airplanes with Lycoming IO-360-B1F6 and AEIO-360-B1G6 engines.

NOTE 2. All placards specified in FAA Approved Airplane Flight Manual must be displayed in the airplane.

NOTE 3. Reserved.

NOTE 4. Reference CAR 04.61 Dated May 31, 1938, to determine propeller diameter and static r.p.m. limits.

NOTE 5. Serial Nos. 0501 and 0502 are modified by installation of the Lycoming O-320-E2A engine per STC SA941CE; and Cleveland 20-80 wheels and brakes, Cleveland IO-5 master cylinders, and Scott tail wheel per STC SA942CE. These airplanes eligible for certification as 2T- 1A-1 airplanes when Identification Plate 00120-1 and FAA Approved Airplane Flight Manual for the 2T-1A-1 dated November 9, 1973 with Rev. "B" dated April 22, 1975, or later FAA approved revision are installed.

NOTE 6. The certification basis for these airplanes does not contain operating categories, i.e., normal, utility or acrobatic, as listed in current certification regulations. However, since certain of these models were designed to perform and have demonstrated the capability to perform the acrobatic maneuvers listed in the approved airplane flight manual plus those maneuvers not exceeding the load factors listed below, the words "acrobatic category" are included so that those persons concerned with these airplanes understand their capacity in terms of current practices. Maneuvering load factors are as follows:

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| A. | 2T-1A-1 | +5.05 to -2.05 |
| B. | 2T-1A-2 | +5.40 to -4.00 |

For the above reasons, the airworthiness certificate for 2T-1A-1 and 2T-1A-2 airplanes are to be issued in the acrobatic category.

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